

Serial No. Not Yet Assigned
Atty. Doc. No. 2003P00692WOUS

Amendments To The Specification:

In the English translation document, please delete the term --Description-- at page 1 written line 1, before the title.

In the English translation document, please add the paragraph at page 1 line 4, after the title, as follows:

--CROSS REFERENCE TO RELATED APPLICATION

This application is the US National Stage of International Application No. PCT/EP2004/005754, filed May 27, 2004 and claims the benefit thereof. The International Application claims the benefits of European Patent application No. 03013648.5 EP filed June 16, 2003, both of the applications are incorporated by reference herein in their entirety.--

In the English translation document, please add the section heading at page 1 line 4, after the newly added CROSS REFERENCE TO RELATED APPLICATION section with the new section heading, as follows:

--FIELD OF THE INVENTION--

In the English translation document, please insert the section heading at page 1 line 13, as follows:

--BACKGROUND OF THE INVENTION--

In the English translation document, please insert the following paragraphs at page 2 line 6, as follows:

In addition, DE 39 09 606 A1 discloses a rotor for an aircraft gas turbine through which a fluid flow can flow. In order to achieve a minimum loss of efficiency of the powerplant cycle, the rotor is either heated or cooled with the fluid during operation. During operation of the gas turbine, compressor inlet air is extracted from the compressor for cooling the rotor and is directed in the interior of the rotor along a hollow shaft and is discharged at the outlet of the turbine. To heat the rotor, compressed and thus already heated compressor air is extracted from the compressor downstream of the first compressor stage and is fed via an external valve to a

mixing chamber in which the heated and compressed air mixes with the cool compressor inlet air. Hot air then flows through the cooler rotor.

In addition, US 6,382,903 discloses the brief heating of the rotor of a gas turbine after it has been started in order to heat the rotor more quickly. To this end, the final compressor air heated by compression is fed to the rotor.

In the English translation document, please insert the section heading at page 3 line 28, as follows:

--SUMMARY OF THE INVENTION—

In the English translation document, please insert the section heading at page 9 line 6, as follows:

--BRIEF DESCRIPTION OF THE DRAWINGS—

In the English translation document, please insert the section heading at page 9 line 21, as follows:

--DETAILED DESCRIPTION OF THE INVENTION--